 CZECH SPORT AIRCRAFT	<b>PS-28 Cruiser</b>	<b>Q-O-04AM</b>	CENTRAL WING SECTION S/N:	<b>602</b>
	Output control	Weight & Balance record	AIRCRAFT S/N:	<b>C0602</b>

**AIRCRAFT SPECIFICATION:**

AIRCRAFT TYPE: **PS-28 Cruiser**  
AIRCRAFT S/N: **C0602**  
CENTRAL WING SECTION S/N: **602**

**INSTALLED EQUIPMENT:**

Rotax 912 ULS2 with airbox and thermostats  
Sensenich 3B0R5R68C three-blade ground adjustable propeller  
ASI, ALT, VSI, Vertical card magnetic compass, Electric turn coordinator  
Electric attitude indicator, Electric directional gyro, MD200-306 CDI  
Garmin GNC255A transceiver, PS Engineering PM3000 intercom  
Garmin GTX328 transponder, Sandia SAE5-35 altitude encoder  
Garmin Aera500 GPS, King AK451 ELT, Antennas  
Engine RPM indicator, Oil pressure and temperature gauges, CHT indicator  
Fuel pressure and quantity indicators, Voltmeter, Engine hours counter  
G-205 trim control and PTT on the control sticks, Trims and flaps electrically actuated  
Kuntzleman LED Landing light in cowl, Cockpit light, Instrument lighting  
AVE-WPST wing tip LED strobe/nav lights, 12V socket  
Adjustable pedals, Dual hydraulic brakes, Parking brake  
Efficient cabin heating, Carburetor preheating, Wheel fairings tricycle  
Leather upholstery, Paint, Sunshade, Arm supports, Stall warning system

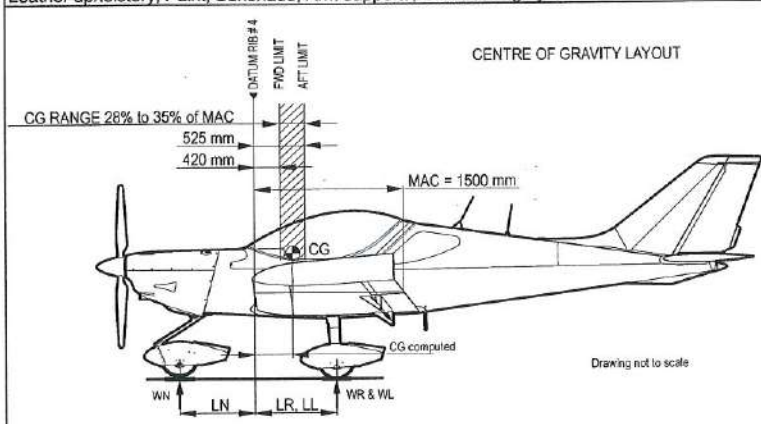
**NOTE:** Data in blue boxes can be changed.

**MEASURING THE AIRCRAFT:**

Inflate the tires, drain the fuel from wing tanks.  
Level the upper fuselage longeron with a spirit level.  
**DATUM** is on leading edge at RIB # 4.  
Measure wheel axle location from **DATUM**.  
Positive arm is behind the **DATUM**.

**LIQUIDS SPECIFICATION:**

FUEL WEIGHT	0.72	KG/LITER
VOLUME (MAX)	114.00	LITRES
OIL WEIGHT	0.90	KG/LITRE
VOLUME (MAX)	3.80	LITRES
COOLANT WEIGHT	1.03	KG/LITER
VOLUME (MAX)	2.50	LITRES



OPERATING C.G. LIMITS :	MM	% MAC
MAC :	1 500.0	100.0
FORWARD CG :	420.0	28.0
REARWARD CG :	525.0	35.0
CG RANGE :	105.0	7.0

EMPTY WEIGHT C.G. LIMITS :	MM	% MAC
MAC :	1 500.0	100.0
FORWARD CG :	427.5	28.5
REARWARD CG :	442.5	29.5
CG RANGE :	15.0	1.0

**AIRCRAFT EMPTY WEIGHT CENTER OF GRAVITY**

POSITION OF:	ARM	MM
RIGHT MAIN WHEEL (LR)	796	
LEFT MAIN WHEEL (LL)	786	
NOSE WHEEL (LN)	-713	
DISTANCE BETWEEN NOSE AND MAIN WHEELS		1 504

*Note: NEGATIVE ARM (-)*

WEIGHING POINT:	SCALE READING	TARE	NET WEIGHT
RIGHT WHEEL (WR)	146.8	0.0	146.8
LEFT WHEEL (WL)	147.6	0.0	147.6
NOSE WHEEL (WN)	92.6	0.0	92.6

*Note: EMPTY WEIGHT IS INCLUDING OIL, COOLANT, HYDRAULIC FLUID AND UNUSUAL FUEL*

**EMPTY WEIGHT:** **387.0** KG

**MAXIMUM TAKEOFF WEIGHT (CS-LSA / ASTM LSA CATEGORY):** **600.0** KG

**MAXIMUM USEFUL WEIGHT:** **213.0** KG

*Note: MAX. USEFUL WEIGHT IS INCLUDING PILOT, PASSENGER, BAGGAGE AND FUEL*

**AIRCRAFT EMPTY WEIGHT C.G. CALCULATION:**

MOMENT (KG MM) = WEIGHT (KG) x ARM (MM)

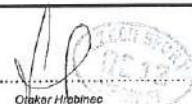
	WEIGHT	ARM	MOMENT
	KG	MM	KG MM
RIGHT MAIN WHEEL	146.8	796	116 852.8
LEFT MAIN WHEEL	147.6	786	116 013.6
NOSE WHEEL	92.6	-713	-66 023.8
<b>TOTAL:</b>	<b>387.0</b>		<b>166 842.6</b>

AIRCRAFT C.G. =  $\frac{\text{TOTAL MOMENT}}{\text{TOTAL WEIGHT}}$  (MM) x  $\frac{100}{\text{MAC}}$  (%)

**RESULT OF AIRCRAFT EMPTY WEIGHT C.G.:** **431.1** MM  
**28.7%** of MAC

DATE: 2017-06-23

ENTITLED PERSON SIGNATURE:



Otakar Hrabinec